

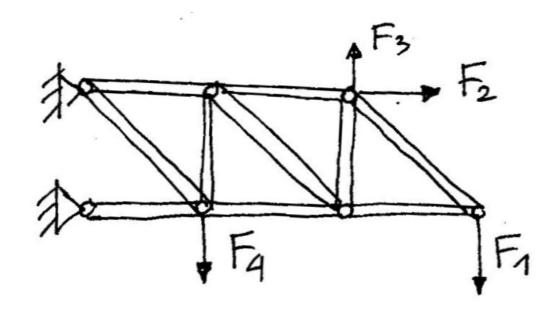
Institute of Aeronautics and Applied Mechanics

# Finite element method (FEM)

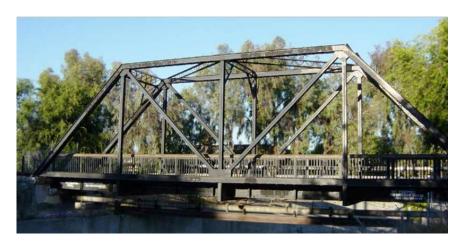
2D truss finite element

04.2021

Trusses - structures made of axial straight ban joined at nodes. Only tensile and compressive forces in members are considered - other internal forces are excluded because at joints are treated as articulated joints.



# Examples of trusses



Bridge



Tower crane

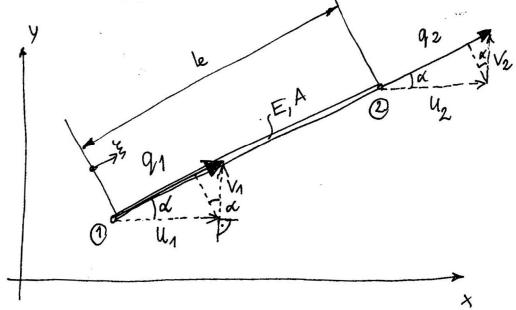


Fuselage



Roof truss

# 2D TRUSS ELEMENT

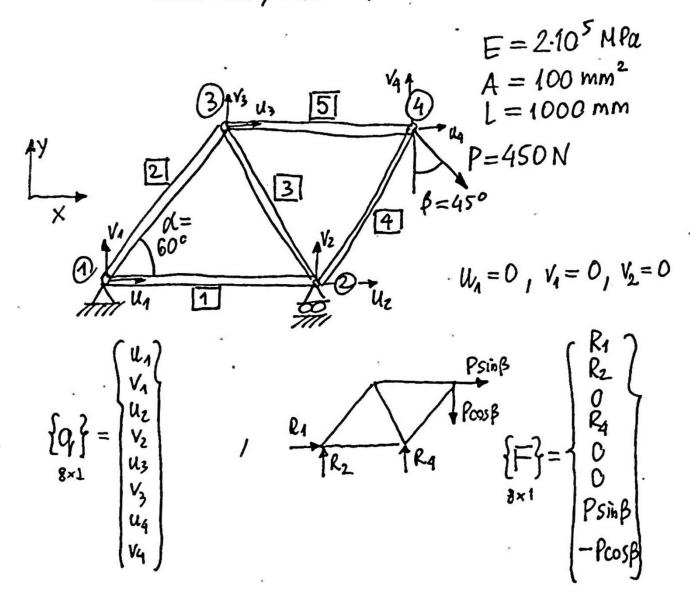


$$\begin{cases}
91 \\
92 \\
e \end{aligned} = \begin{bmatrix}
C & S & O & O \\
O & O & C & S
\end{bmatrix} \cdot \begin{cases}
U_1 \\
U_2 \\
V_2 \\
V_3 \\
V_4 \\
V$$

elastic strain energy.

$$[K_g]_e = \frac{EA}{k} \begin{bmatrix} c^2 & sc - c^2 - sc \\ sc & s^2 - sc & -s^2 \\ -c^2 - sc & c^2 & sc \\ -sc - s^2 & sc & s^2 \end{bmatrix}$$

EXAMPLE . BUILD A FE NODEL OF A 2-D TRUSS. FIND NODEL DISPLACEMENTS, STRESSES, INTERNAL FORCES AND REACTIONS



$$\begin{bmatrix} 2 \\ 3 \\ 4x4 \end{bmatrix}^{\frac{1}{3}} = \begin{bmatrix} 4 \\ 4 \\ 4x4 \end{bmatrix}^{\frac{1}{3}} = \begin{bmatrix} 4 \\ 4x$$

$$\begin{bmatrix} K \end{bmatrix} = \sum_{e=1}^{5} \begin{bmatrix} 5\sqrt{3} - 4 & 0 & -1 & -\sqrt{3} & 0 & 0 \\ 3 & 3 & 0 & 0 & -\sqrt{3} & -3 & 0 & 0 \\ 4 & 0 & 6 & 0 & -1 & 3 & -1 & -\sqrt{3} \\ 0 & 0 & 6 & 3 & -3 & -\sqrt{3} & -3 \\ -1 - 6 - 1 & 3 & 6 & 0 & -4 & 0 \\ -13 - 3 & 3 - 3 & 0 & 6 & 0 & 0 \\ 0 & 0 - 1 - 6 - 4 & 0 & 5 & \sqrt{3} \\ 0 & 0 - 6 - 3 & 0 & 0 & 3 & 3 \end{bmatrix}$$

$$\begin{bmatrix} K \end{bmatrix} \cdot \begin{Bmatrix} q \end{Bmatrix} = \begin{Bmatrix} F \end{Bmatrix}$$

- + boundary conditions:  $U_1=0$ ,  $V_2=0$ 

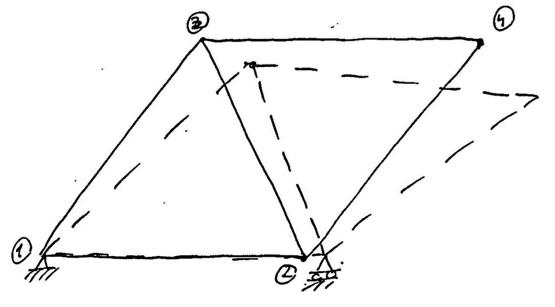
$$U_2 = 0.3362 \cdot 10^{-2} \text{ mm}$$

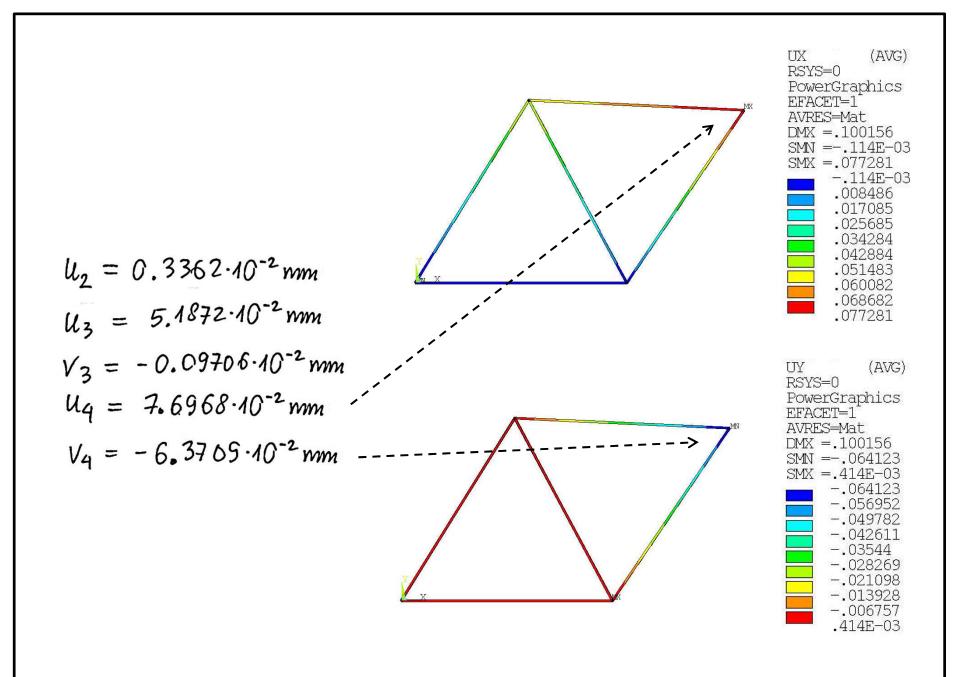
$$U_3 = 5.1872 \cdot 10^{-2} \text{ mm}$$

$$V_3 = -0.09706 \cdot 10^{-2} \text{ mm}$$

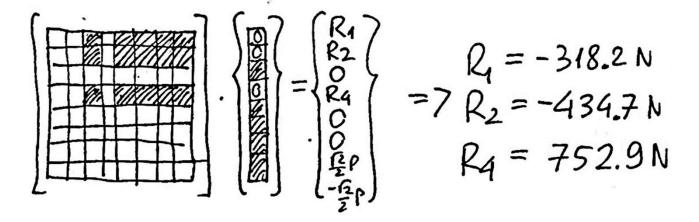
$$U_4 = 7.6968 \cdot 10^{-2} \text{ mm}$$

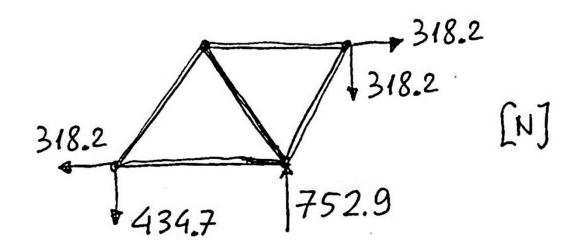
$$V_4 = -6.3709 \cdot 10^{-2} \text{ mm}$$





## reactions





stresses and internal forces:

$$\begin{bmatrix} 1 \end{bmatrix} \begin{cases} 91/3 \\ q_2 \end{bmatrix}_1 = \begin{bmatrix} T_t \\ 1 \end{cases} \cdot \begin{cases} 99/3 \\ 4\times 1 \end{bmatrix} = \begin{bmatrix} C_1 & S_1 & 0 & G \\ 0 & C_1 & S_1 \end{bmatrix} \cdot \begin{cases} V_1 \\ V_2 \\ V_2 \end{pmatrix}_1 = \begin{cases} 0 \\ 0.33622 \cdot 10^{-2} \end{cases}$$

$$G_1 = \begin{bmatrix} E \\ (q_2 - q_1)_1 = 0.67 \text{ Mfa} \\ V_2 \end{bmatrix} = \begin{bmatrix} V_1 \\ (4\times 1) \end{bmatrix} = \begin{bmatrix} C_2 & S_2 & 0 & C \\ 2\times 1 \end{bmatrix} \cdot \begin{cases} q_1 \\ q_2 \end{bmatrix}_2 = \begin{bmatrix} T_t \\ 2\times 1 \end{bmatrix}_2 \cdot \begin{cases} q_2 \\ q_2 \end{bmatrix}_2 = \begin{bmatrix} C_2 & S_2 & 0 & C \\ C & 0 & C_2 & S_2 \end{bmatrix} \cdot \begin{cases} V_1 \\ V_2 \\ V_3 \end{pmatrix}_2 = \begin{cases} 0 \\ 2.5096 \cdot 10^{-2} \end{cases}$$

$$G_2 = \begin{bmatrix} E \\ (q_2 - q_1)_2 = 5.02 \text{ Mfa} \\ V_2 \end{bmatrix} \quad N_2 = G_2 \cdot A = 502 \text{ N}$$

$$\begin{bmatrix} q_1 \\ q_2 \end{bmatrix}_3 = \begin{bmatrix} T_t \\ q_2 \end{bmatrix}_3 \cdot \begin{cases} q_9 \\ q_3 \end{bmatrix}_3 = \begin{bmatrix} C_3 & S_3 & 0 & C \\ C & 0 & C_3 & S_3 \end{bmatrix} \cdot \begin{cases} V_2 \\ V_3 \\ V_3 \end{cases} = \begin{cases} -0.16811 \cdot 10^{-2} \\ -2.67766 \cdot 10^{-2} \end{cases}$$

$$G_3 = \begin{bmatrix} E \\ (q_2 - q_1)_3 = -5.02 \text{ Mfa} \end{cases} \quad N_3 = G_3 \cdot A = -502 \text{ N}$$

$$(possible buckling)$$

$$\begin{aligned}
& \left[ \frac{91}{92} \right]_{4} = \left[ \frac{1}{4} \cdot \left[ \frac{99}{9} \right]_{4} = \left[ \frac{1}{6} \cdot \frac{99}{6} \cdot \frac{1}{6} \cdot$$

### Stress [MPa]

